



Airworthiness Directive

AD No.: 2018-0202R1

Issued: 25 September 2018

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

ROLLS-ROYCE plc

Type/Model designation(s):

RB211-535E4 engines

Effective Date: 25 September 2018 (same as original issue)

TCDS Number(s): EASA E.061

Foreign AD: Not applicable

Revision: This AD revises EASA AD 2018-0202 dated 11 September 2018, which superseded United Kingdom (UK) CAA AD 002-01-2000 dated January 2000.

ATA 72 – Engine – Low Pressure Compressor Blades – Inspection

Manufacturer(s):

Rolls-Royce plc (RR)

Applicability:

RB211-535E4-37, RB211-535E4-B-37, RB211-535E4-C-37 and RB211-535E4-B-75 engines, all serial numbers.

These engines are known to be installed on, but not limited to, Boeing 757 and Tupolev TU-204 aeroplanes.

Definitions:

For the purpose of this AD, the following definitions apply:

The NMSB: RR Alert Non-Modification Service Bulletin (NMSB) RB.211-72-AC879 Revision 9 dated 23 April 2018. The NMSB has an 'A' (Alert) in the number, but an earlier or later revision may not have that 'A'. This kind of change does not effectively alter the publication references for the purpose of this AD.

The modification SB: RR Service Bulletin (SB) RB.211-72-C946 (any revision).



Affected fan blade: All fan blades, all Part Numbers (P/N), except those that are post-SB/post-mod 72-C946, which can be visually identified by a blue triangle etched on the blade aerofoil suction surface close to the leading edge tip.

Reason:

During engine overhaul, inspection of a high life set of low pressure compressor (LPC) fan blades revealed small cracks in the blade roots on the concave root flank. These cracks had originated at the edge of bedding from multiple origins. Root cause analysis indicated the cause of the crack initiation to be the absence of the anti-fretage coating.

This condition, if not detected and corrected, could lead to fan blade failure, possibly resulting in release of high energy non-contained debris from the engine, with consequent damage to the aeroplane.

To address this condition, RR issued NMSB RB.211-72-AC879 (original issue, later revised), providing instructions to inspect high life blades, either on-wing or during engine overhaul. Depending on flight profile flown, different inspection intervals were introduced. Consequently, the UK CAA classified that NMSB as mandatory and issued AD 002-01-2000 accordingly, requiring those repetitive inspections.

Since that AD was issued, it was reported that some engines were operated outside the profiles initially specified, and new flight profiles were introduced to mitigate the risk of overflying the recommended flight profiles. Consequently, the inspection intervals were extended for engines operating within RB211-535E4-B-37 flight profiles C, D and E, and RR issued the NMSB accordingly. Additionally, RR introduced inspection instructions for engines operating within RB211-535E4-C-37 flight profile F and RB211-535E4-37 flight profile G in the NMSB.

For the reasons described above, EASA issued AD 2018-0202, retaining the requirements of UK CAA AD 002-01-2000, which was superseded, amending the compliance times and adding repetitive inspections for RB211-535E4-37, RB211-535E4-B-37 and RB211-535E4-C-37 engines operating within flight profiles C, D, E, F and G. That AD also provided a modification as optional terminating action for the repetitive inspections.

This AD is revised to correct paragraph (1), indicating that only affected fan blades must be inspected.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Inspection(s):

- (1) Before exceeding the threshold as indicated in Table 1 of this AD, or within 30 days after the effective date of this AD, whichever occurs later, and thereafter, at intervals not to exceed the applicable value as specified in Table 1 of this AD, depending on the engine model and operating flight profile (see Note 1 of this AD), and on the inspection method (root probe, or surface probe), inspect each affected fan blade in accordance with the instructions of the NMSB.



Note 1: The 'threshold' values specified in Table 1 of this AD are engine flight cycles (EFC) accumulated since first installation of an affected fan blade on an engine.

Table 1 – Fan Blade Inspection (see Note 1 of this AD)

Model / Flight Profile(s)	Threshold	Inspection (Probe) Method / Interval (not to exceed)	
		Root	Surface Wave
535E4-37 / B and G	15 000 EFC	850 EFC	700 EFC
535E4-C-37 / F			
535E4-B-37 / E and C	20 000 EFC	1 200 EFC	1 000 EFC
535E4-B-75 / all			
535E4-37 / A			
535E4-B-37 / D			
		1 400 EFC	1 150 EFC
		1 500 EFC	1 200 EFC

Note 2: For intermix operation between 535E4 flight profiles A, B and G, 535E4-B flight profiles C, D and E, and 535E4-C flight profile F, the currently used flight profile can be applied to determine the applicable inspection threshold and repeat interval as specified in Table 1 of this AD.

- (2) At the point of changing flight profiles, in accordance with the instructions of the NMSB, inspect the affected fan blades before exceeding the new flight profile threshold or cyclic interval, or within 200 EFC after changing flight profiles, whichever occurs later, without exceeding the previous flight profile threshold or interval.

Recalculation:

- (3) For an engine having an affected fan blade installed that has operated in more than one flight profile or engine rating at any time during its service life, before exceeding 15 000 EFC, or within 200 EFC after the effective date of this AD, whichever occurs later, and thereafter, within 200 EFC after each change of flight profile, without exceeding the previous flight profile threshold or interval, recalculate the equivalent cycles since new in accordance with the instructions of the NMSB.

Corrective Action(s):

- (4) If, during any inspection as required by paragraph (1) or (2) of this AD, as applicable, any crack indication is found that exceeds the criteria as specified in the NMSB, before next flight, or before release to service of the engine, as applicable, replace the blade with a serviceable blade in accordance with the instructions of the NMSB.

Credit:

- (5) Inspections and corrective action(s) on an engine, accomplished before the effective date of this AD in accordance with the instructions of NMSB RB.211-72-C879 at Revision 8 or any earlier revision, are acceptable to comply with the initial requirements of this AD for that engine. After the effective date of this AD, the instructions of the NMSB (Revision 9) must be used.



Terminating Action:

- (6) Modification of an engine in accordance with the modification SB constitutes terminating action for the repetitive inspections required by this AD for that engine.

Parts Installation:

- (7) From the effective date of this AD, it is allowed to install on any engine an affected fan blade, provided that the part is new (never installed on an engine); or has not exceeded the applicable threshold as indicated in Table 1 of this AD; or has passed an inspection in accordance with the instructions of the NMSB; as applicable, and that, following installation, the affected fan blade is inspected as required by this AD.

Ref. Publications:

Rolls-Royce NMSB RB.211-72-C879 original issue dated 11 January 2000, or Revision 1 dated 30 June 2000, or Revision 2 dated 4 January 2002, or Revision 3 dated 9 October 2002, or Revision 4 dated 2 April 2004, or Revision 5 dated 8 March 2007, or Revision 6 dated 14 December 2007, or Revision 7 dated 1 May 2009, or Revision 8 dated 18 November 2015, or Alert NMSB RB.211-72-AC879 Revision 9, dated 23 April 2018.

Rolls-Royce SB RB.211-72-C946 original issue dated 29 September 2000, or Revision 1 dated 28 August 2001, or Revision 2 dated 26 September 2002, or Revision 3 dated 22 September 2008, or Revision 4 dated 22 June 2010.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 04 May 2018 as PAD 18-066 for consultation until 01 June 2018. The Comment Response Document can be found in the [EASA Safety Publications Tool](#), in the compressed (zipped) file attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#).
5. For any question concerning the technical content of the requirements in this AD, please contact: your designated Rolls-Royce representative, or download the publication from your Rolls-Royce Care account at <https://customers.rolls-royce.com>.



If you do not have a designated representative or Rolls-Royce Care account, please contact **Corporate Communications** at **Rolls-Royce plc**, P.O. Box 31, Derby, DE24 8BJ, United Kingdom, telephone +44 (0)1332 242424, or

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