Airworthiness Directive

AD No.: 2020-0019

Issued: 05 February 2020

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS

Type/Model designation(s):

A318, A319, A320 and A321 aeroplanes

Effective Date: 19 February 2020

TCDS Number(s): EASA.A.064

Foreign AD: Not applicable

Supersedure: None

ATA 53 – Fuselage – Lateral Window Frame Upper Fastener Holes – Inspection

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Manufacturer(s):

Airbus, formerly Airbus Industrie

Applicability:

Airbus A318-111, A318-112, A318-121, A318-122, A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132, A319-133, A320-211, A320-212, A320-214, A320-215, A320-216, A320-231, A320-232, A320-233, A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231 and A321-232 aeroplanes, all manufacturer serial numbers, except:

- aeroplanes on which Airbus modification (mod) 158423 was embodied in production;
- A318, A319 and A320 aeroplanes on which Airbus mod 160000 was embodied in production;
- A319 and A320 aeroplanes on which Airbus Service Bulletin (SB) A320-57-1193 was embodied in service;
- A321 aeroplanes on which Airbus mod 160021 was embodied in production;
- A318 aeroplanes on which Airbus mod 39195 was embodied in production, or Airbus SB A320-00-1219 was embodied in service; and
- A319 aeroplanes on which Airbus mod 28238, mod 28162 and mod 28342 were embodied in production.

Definitions:

For the purpose of this AD, the following definitions apply:

The SBs: Airbus SB A320-53-1448 and Airbus SB A320-53-1449.

The modification SBs: Airbus SB A320-53-1335 and Airbus SB A320-53-1336.

Affected area: Central/lateral window frame junction upper area around fastener holes, right-hand side and left-hand side.

Reason:

During the accomplishment of the Airworthiness Limitation Items (ALI) task 531125 (Detailed inspection of central/lateral window frame junction upper area around fastener holes) on aeroplanes in service, and during full scale fatigue tests, crack findings have been reported in fastener holes which were not subject to the detailed inspection (DET).

This condition, if not detected and corrected, could reduce the structural integrity of the aeroplane.

To address this possible unsafe condition, Airbus issued the inspection SBs to redefine the area to be inspected, the inspection means (DET and special detailed inspection (SDI)) and the instructions. Airbus also issued the modifications SBs, providing instructions to replace the sliding window frame with new, and revised the Airworthiness Limitation Section (ALS) Part 2 (now at revision 8), removing, among other updates, the ALI task 531125. Using ALS Part 2 revision 8 is acceptable for compliance with the requirements of EASA AD 2018-0288, which requires accomplishment of the actions specified in the ALS Part 2.

For the reasons described above, this AD requires repetitive DET and/or SDI of the affected area and fastener holes, depending on aeroplane configuration, and, depending on findings, accomplishment of applicable corrective action(s).

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Inspection(s):

(1) Before exceeding 48 000 flight cycles (FC) since aeroplane first flight, or since replacement of the sliding window frame with new one in accordance with the modification SBs or in accordance with an Airbus Repair Design Approval Sheet (RDAS), as applicable, or within 15 000 FC after last accomplishment of ALI task 531125 before the effective date of this AD, or within 3 months after the effective date of this AD, whichever occurs later, and, thereafter, at intervals not to exceed 29 800 FC, inspect each affected area and fastener holes in accordance with the instructions of the SBs, as applicable.

Corrective Action(s):

(2) If, during any inspection as required by paragraph (1) of this AD, discrepancies are detected, before next flight, contact Airbus for approved instructions and accomplish those instructions accordingly.

Terminating Action(s):

(3) None.

Ref. Publications:

Airbus SB A320-53-1335 original issue dated 12 March 2019.

Airbus SB A320-53-1336 original issue dated 12 March 2019.

Airbus SB A320-53-1448 original issue dated 05 August 2019.

Airbus SB A320-53-1449 original issue dated 05 August 2019.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD This Proposed AD will be closed for consultation on 08 October 2019.

2. This AD was posted on 10 September 2019 as PAD 19-169 for consultation until 08 October 2019. The Comment Response Documents can be found in the EASA Safety Publications Tool, in the compressed (zipped) file attached to the record for this AD.

3. Enquiries regarding this AD should be referred to the EASA Programming and Continued Airworthiness Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.

4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the EU aviation safety reporting system.

5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – EIAS; Fax +33 5 61 93 44 51;
E-mail: account.airworth-eas@airbus.com.