



Airworthiness Directive

AD No.: 2021-0017

Issued: 14 January 2021

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS

Type/Model designation(s):

A340 aeroplanes

Effective Date: 28 January 2021

TCDS Number(s): EASA.A.015

Foreign AD: Not applicable

Supersedure: This AD supersedes DGAC France AD 98-026-079(B) dated 28 January 1998.

ATA 53 – Fuselage – Type A Emergency Exit Door Frames – Modification / Reinforcement

Manufacturer(s):

Airbus, formerly Airbus Industrie

Applicability:

Airbus A340-211, A340-213, A340-311 and A340-312 aeroplanes, manufacturer serial numbers 0002 to 0005 inclusive, 007, 029, 048, 078 and 084.

Definitions:

For the purpose of this AD, the following definitions apply:

Affected areas: Frame / crossbeam connections at fuselage frames (FR) 53.3 and FR53.5, on left-hand and right-hand sides.

The SB: Airbus Service Bulletin (SB) A340-53-4053 Revision 02.

Reason:

During the full-scale fatigue testing of the A340 aeroplane, cracks were found in the affected areas.

This condition, if not detected and corrected, could affect the structural integrity of the aeroplane.



To address this potential unsafe condition, Airbus issued SB A340-53-4053 (original issue, later revised) to provide instructions to reinforce the affected areas by installing straps and performing cold working at specific holes. Consequently, DGAC France issued AD 98-026-079(B) to require the modification of the centre fuselage.

Since that AD was issued, Airbus introduced a lower threshold for the modification embodiment, obtained within the frame of the certification of the A340 Extended Service Goal exercise. No additional work is required for aeroplanes already modified in accordance with the instructions of Airbus SB A340-53-4053 (at any Revision). In addition, it was identified that the Applicability of DGAC France AD 98-026-079(B) was incorrect, where too many models were mentioned.

For the reasons described above, this AD retains the requirements of DGAC France AD 98-026-079(B), which is superseded, corrects its Applicability, and reduces the compliance time.

Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

Modification(s):

- (1) Before exceeding 9 420 flight cycles since aeroplane first flight, or within 18 months after the effective date of this AD, whichever occurs later, modify the affected areas in accordance with the instructions of the SB.

Credit:

- (2) Modification of an aeroplane, accomplished before the effective date of this AD in accordance with the instructions of Airbus SB A340-53-4053 at original issue, or Revision 01, is an acceptable method to comply with the requirements of paragraph (1) of this AD for that aeroplane.

Ref. Publications:

Airbus SB A340-53-4053 original issue dated 14 May 1996, or Revision 01 dated 13 August 1997, or Revision 02 dated 11 September 1998.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 14 December 2020 as PAD 20-196 for consultation until 11 January 2021. No comments were received during the consultation period.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on



a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.

5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – IIAL (Airworthiness Office), E-mail: airworthiness.A330-A340@airbus.com.

